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Thin Airfoil Theory - Virginia Tech Given By The Equations: Determine The Aerodynamic Characteristics

$$C_L = \frac{2\pi\alpha}{\pi} = 2\alpha$$

$$C_D = 0$$

ANSYS Workbench Tutorial – Flow Over An Airfoil The First Two Rows Of This Excel File Help Keep The Data Organized, But They Are Not Required. 13. In The First Line Of Data, Change The X Coordinate To “1” And The Y Coordinate To “0”. The First Data Point Should Be Located At The Trailing Edge Of The Airfoil, But Due To Resolution And Accuracy Limits It Is Not Exactly (1,0). 14. Enter “1” In Cell A3 And A4, And Fill Through The ...

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